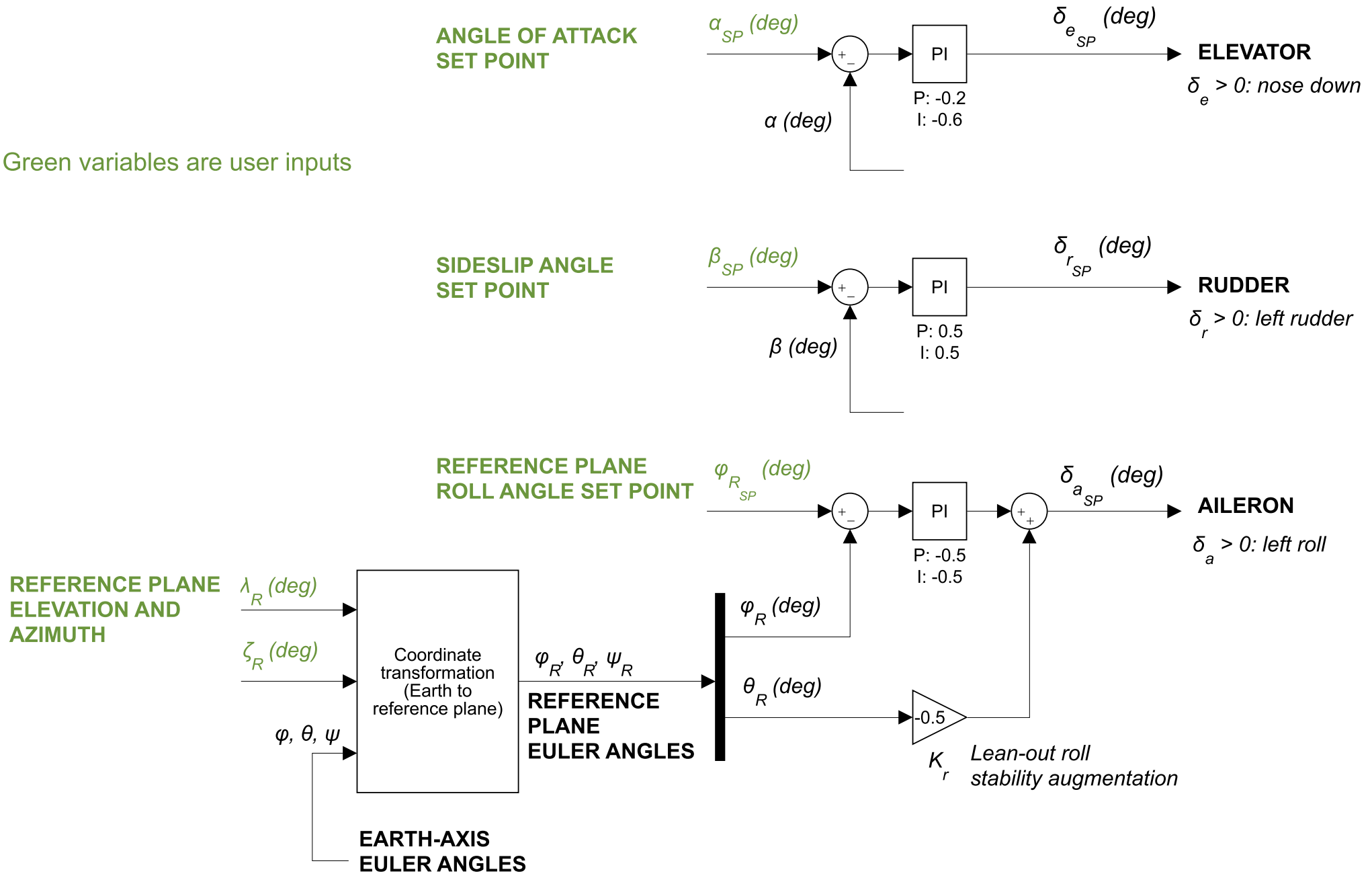
ANGLE OF ATTACK SET POINT
α_{SP} (deg)
α (deg)
PI
P: -0.2
I: -0.6
$\delta_{e_{SP}}$ (deg)
ELEVATOR
δ_e > 0: nose down
Green variables are user inputs
SIDESLIP ANGLE SET POINT
β_{SP} (deg)
β (deg)
PI
P: 0.5
I: 0.5
$\delta_{r_{SP}}$ (deg)
RUDDER
δ_r > 0: left rudder
REFERENCE PLANE ROLL ANGLE SET POINT
$\varphi_{R_{SP}}$ (deg)
PI
P: -0.5
I: -0.5
$\delta_{a_{SP}}$ (deg)
AILERON
δ_a > 0: left roll
REFERENCE PLANE ELEVATION AND AZIMUTH
λ_R (deg)
ζ_R (deg)
φ, θ, ψ
Coordinate transformation (Earth to reference plane)
$\varphi_R, \theta_R, \psi_R$
REFERENCE PLANE EULER ANGLES
φ_R (deg)
θ_R (deg)
-0.5
K_r
Lean-out roll stability augmentation
EARTH-AXIS EULER ANGLES