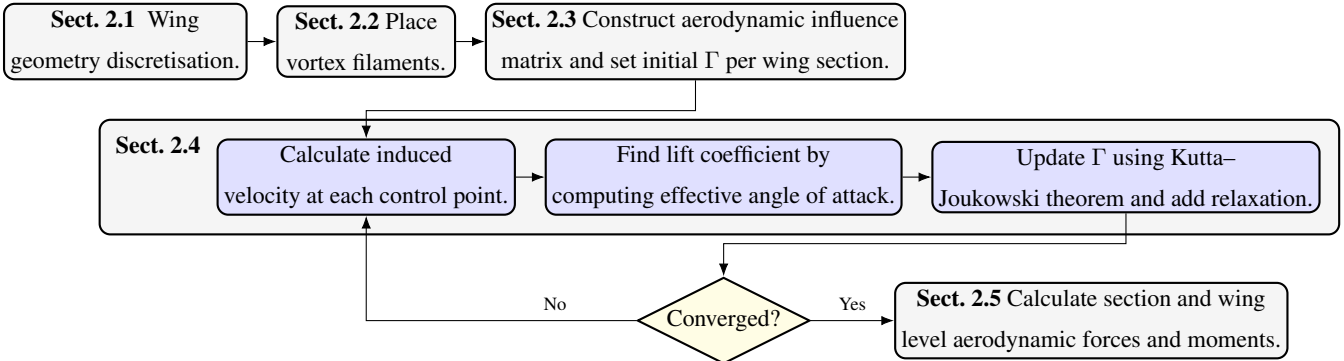

Sect. 2.1 Wing geometry discretisation.
Sect. 2.2 Place vortex filaments.
Sect. 2.3 Construct aerodynamic influence matrix and set initial Γ per wing section.
Sect. 2.4
Calculate induced velocity at each control point.
Find lift coefficient by computing effective angle of attack.
Update Γ using Kutta–Joukowski theorem and add relaxation.
Converged?
No
Yes
Sect. 2.5 Calculate section and wing level aerodynamic forces and moments.